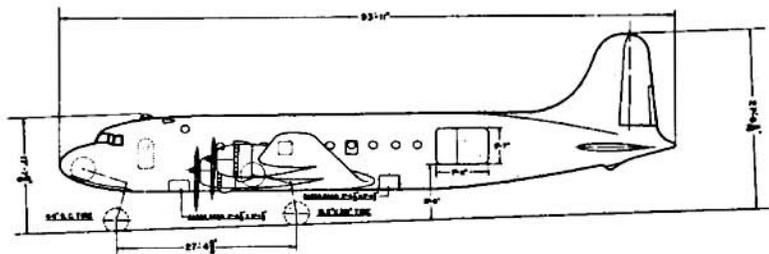
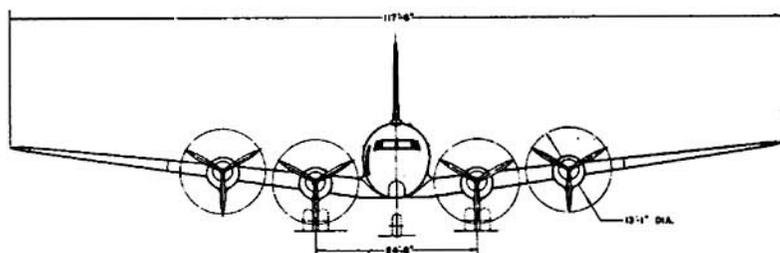
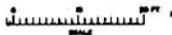
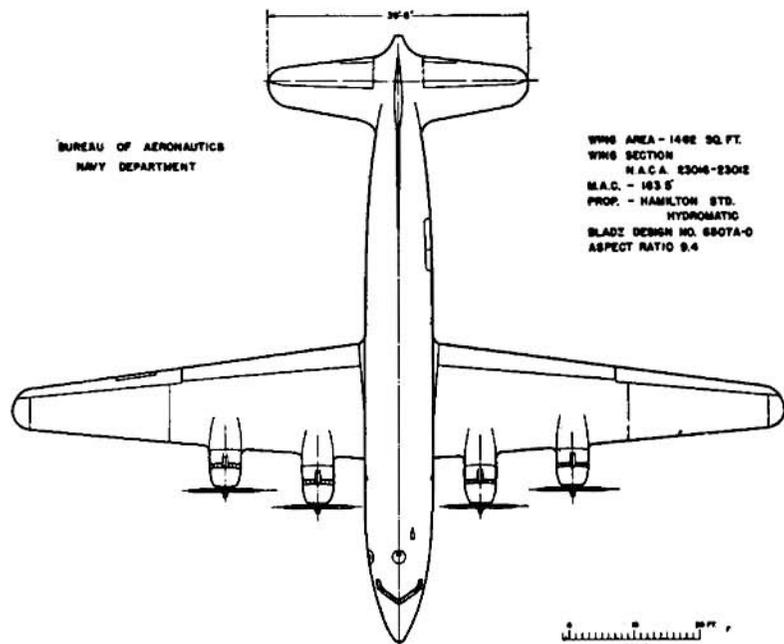


STANDARD AIRCRAFT CHARACTERISTICS  
C-54Q SKYMASTER

BUREAU OF AERONAUTICS  
NAVY DEPARTMENT

WING AREA - 1492 SQ. FT.  
WING SECTION  
N.A.C.A. 23046-23042  
M.A.C. - 163.5'  
PROP. - HAMILTON STD.  
HYDROBATIC  
BLADE DESIGN NO. 6807A-0  
ASPECT RATIO 9.4



DESCRIPTIVE ARRANGEMENT

**POWER PLANT**

NO. & MODEL.....(4) R-2000-9  
 MFR.....Pratt & Whitney  
 SUPERCH.....1 Stage, 2 Speed  
 RED. GEAR RATIO.....2:1  
 PROF. MFR.....Ham. Std.  
 PROF. DES. NO.....6507A-0  
 NO. BLD./DIA.....3/13' -1"

**RATINGS**

BHP • RPM • ALT.

|       |       |       |            |
|-------|-------|-------|------------|
| T.O.  | 1,450 | 2,700 | S.L.       |
| NORM. | 1,100 | 2,550 | S.L.       |
|       |       |       | to 7,500'  |
|       | 1,000 | 2,550 | 10,000'    |
|       |       |       | to 17,000' |

SPEC. NO. A-9021-B

**ACCOMMODATIONS**

Crew.....6  
 Troop Seats.....50  
 Litters.....36  
 Max. Floor Load..200 lbs/sq.ft.  
 Boom Hoist Cap.....2,000 lbs.  
 Cargo Door  
 Height.....67"  
 Width.....95"  
 Max. Pay Load.....29,000 lbs.

**MISSION AND DESCRIPTION**

The Douglas "Skymaster" R5D-3 troop and cargo transport is the same as the Air Force C54-D.

It is of conventional structure, with a 3 spar center section and single spar wing. It is fitted with slotted flaps and tricycle landing gear. Power plants are demountable and interchangeable. Deicer shoes may be fitted to wing, fin and stabilizer. The glider tow fittings have been removed from Navy planes.

The design of the R5D-3 stems from the commercial DC-4 design.

**DEVELOPMENT**

Service use.....1945

**DIMENSIONS**

WING  
 AREA.....1,462 sq.ft.  
 SPAN.....117' -6"  
 M. A. C.....13' -7"  
 LENGTH.....93' -11"  
 HEIGHT.....27' -6"  
 TREAD.....24' -8"  
 FROE. GRD. CLEARANCE....3' -0"

**WEIGHTS**

| <u>LOADINGS</u> | <u>LBS</u>  | <u>L.F.</u> |
|-----------------|-------------|-------------|
| EMPTY.....      | 40,310..... |             |
| BASIC.....      | 41,272..... |             |
| DESIGN.....     | 62,000..... | 2.5.        |
| MAX. T.O.....   | 73,000..... |             |
| MAX. LDG.....   | 66,000..... |             |

L.F. varies from 2.33 to 2.6 depending on the amount of fuel in the wings. See flight handbook.  
 All weights are actual.

**FUEL AND OIL**

| <u>No. Tanks</u>        | <u>Tot. Gals.</u> | <u>Location</u> |
|-------------------------|-------------------|-----------------|
| 6                       | 2,820             | Wing            |
| 2                       | 900               | Fuselage        |
| FUEL GRADE.....100/133  |                   |                 |
| FUEL SPEC....MIL-F-5572 |                   |                 |

**OIL**

CAPACITY (Gals).....138  
 GRADE.....1100  
 SPEC.....MIL-L-6082

**ELECTRONICS**

LF & MHF COMMAND...SCR-274-N  
 VHF COMMAND.....AN/ARC-1  
 LIAISON.....(2) AN/ART-13  
 IFF.....AN/AFX-6  
 COMPASS.....SCR-269-G  
 and AN/ARN-7  
 MARKER BEACON.....AN/ARN-8  
 ALTIMETER.....SCR-718-A  
 and AN/AFN-1  
 LORAN.....AN/AFN-4

(Cont'd on NOTES page)

## PERFORMANCE SUMMARY

| TAKE-OFF LOADING CONDITION          |            | CARGO      |  |  |  |
|-------------------------------------|------------|------------|--|--|--|
|                                     |            | FULL FUEL  |  |  |  |
| TAKE-OFF WEIGHT                     | lb.        | 73,000     |  |  |  |
| Fuel                                | lb.        | 22,320     |  |  |  |
| Payload                             | lb.        | 9,408      |  |  |  |
| Wing loading                        | lb./sq.ft. | 49.9       |  |  |  |
| Stall speed - power-off             | kn.        | 79.5       |  |  |  |
| Take-off run at S.L. - calm (B)     | ft.        | 2,700      |  |  |  |
| Take-off run at S.L. 25 kn. wind(B) | ft.        | 1,400      |  |  |  |
| Take-off to clear 50 ft. - calm (B) | ft.        | 4,650      |  |  |  |
| Max. speed/altitude (A)             | kn./ft.    | 243/18,300 |  |  |  |
| Rate of climb at S.L. (A)           | fpm        | 830        |  |  |  |
| Time: S.L. to 10,000 ft. (A)        | min.       | 14.6       |  |  |  |
| Time: S.L. to 20,000 ft. (A)        | min.       | 39.1       |  |  |  |
| Service ceiling (100 fpm)           | ft.        | 22,700     |  |  |  |
| Combat range                        | n.mi.      | 2,740      |  |  |  |
| Average cruising speed              | kn.        | 174        |  |  |  |
| Cruising altitude(s)                | ft.        | 5,000      |  |  |  |
| Combat radius                       | n.mi.      |            |  |  |  |
| Average cruising speed              | kn.        |            |  |  |  |
| COMBAT LOADING CONDITION            |            |            |  |  |  |
| COMBAT WEIGHT                       | lb.        |            |  |  |  |
| Engine power                        |            |            |  |  |  |
| Fuel                                | lb.        |            |  |  |  |
| Combat speed/altitude               | kn./ft.    |            |  |  |  |
| Rate of climb/altitude              | fpm/ft.    |            |  |  |  |
| Combat ceiling (500 fpm)            | ft.        |            |  |  |  |
| Rate of climb at S.L.               | fpm        |            |  |  |  |
| Max. speed at S.L.                  | kn.        |            |  |  |  |
| Max. speed/altitude                 | kn./ft.    |            |  |  |  |
| LANDING WEIGHT                      |            |            |  |  |  |
| Fuel                                | lb.        |            |  |  |  |
| Stall speed - power-off             | kn.        |            |  |  |  |
| Stall speed - with approach power   | kn.        |            |  |  |  |

## NOTES

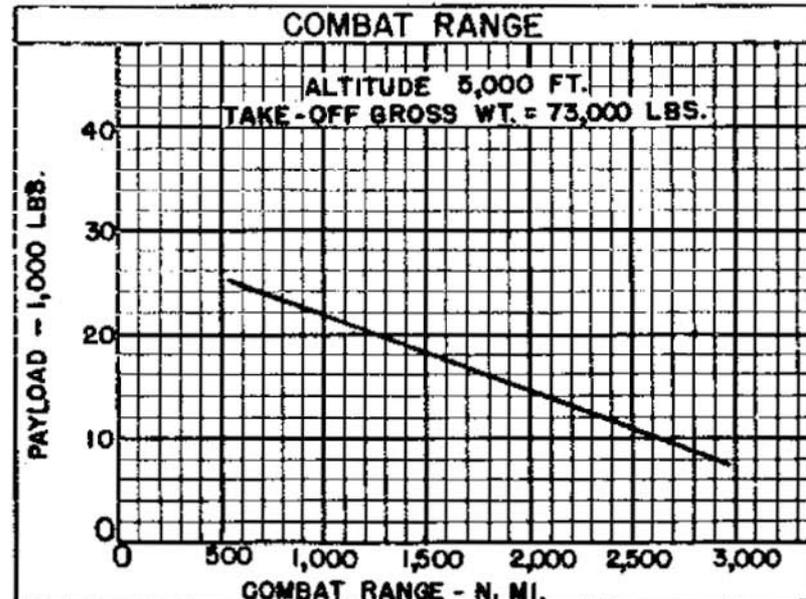
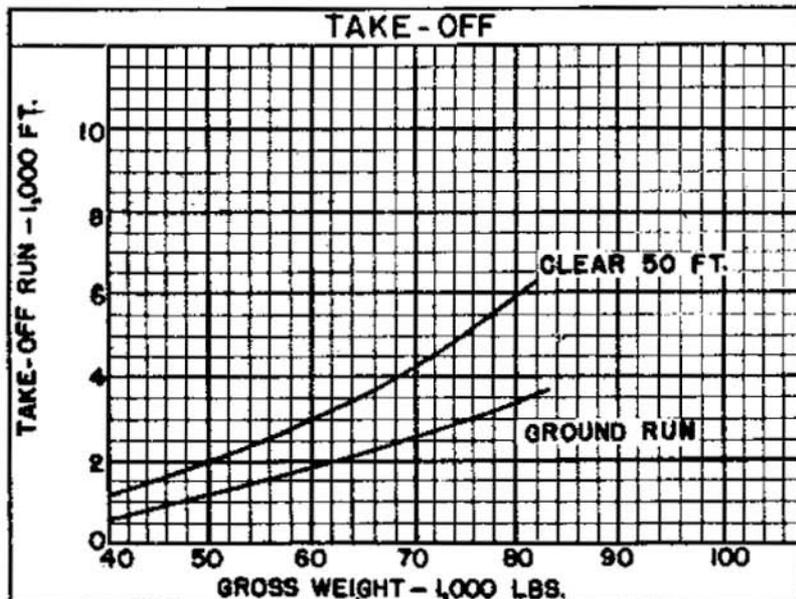
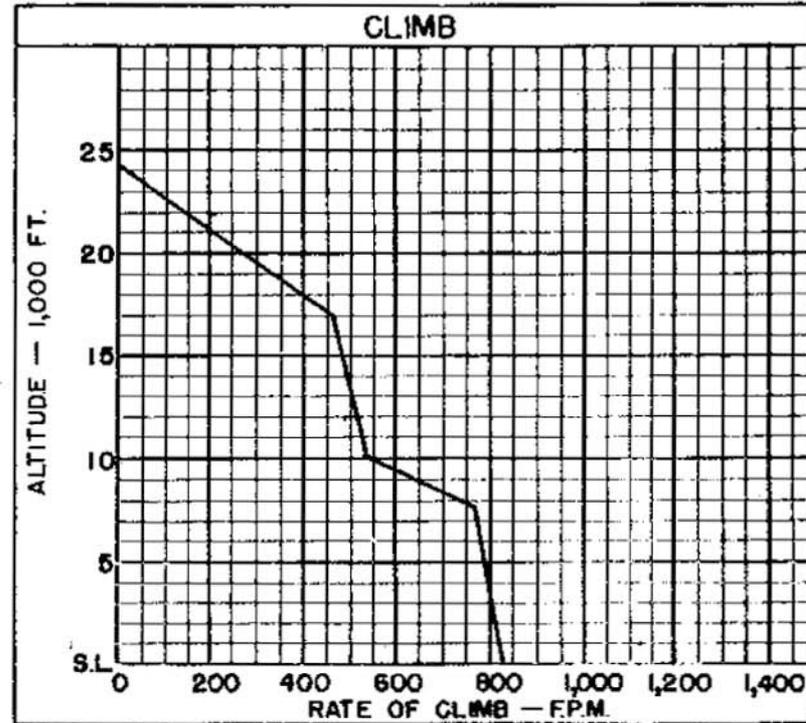
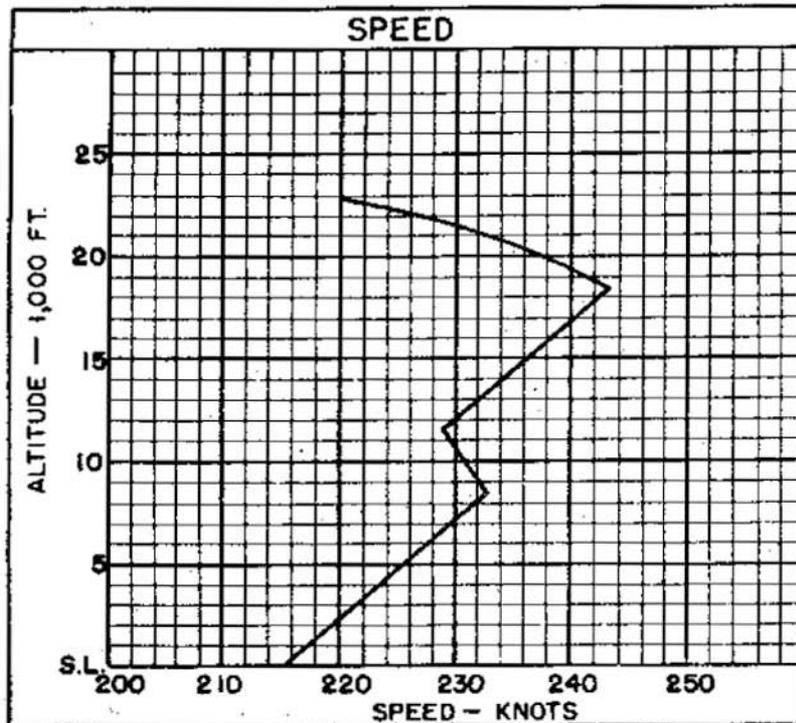
(A) Normal Rated Power

(B) Take-Off Power

Performance Basis: Flight Test of R5D-3 airplane.

Range and radius are based on engine specification fuel consumption data increased by 5%.

Performance includes the effect of deicer boots. Removal of deicer boots increases Vmax @ ACA by 4.5 knots.



# NOTES

Performance with one engine inoperative, flaps and landing gear retracted and propeller feathered is estimated to be:

|                                     |         |        |
|-------------------------------------|---------|--------|
| Gross Weight.....                   | Lbs.    | 73,000 |
| Rate of Climb - S.L., T.O. Power... | Ft/Min. | 620    |
| Service Ceiling - Normal Power..... | Ft.     | 15,400 |

## COMBAT RANGE

WARM-UP, TAKE-OFF, ACCELERATION: 10 minutes at normal rated power.  
 CLIMB: On course to 5,000 feet at normal rated power.  
 CRUISE: At speed for long range at 5,000 feet.  
 RESERVE: 30 minutes at speed for long range plus 5% of initial fuel load.

COMBAT RANGE = CLIMB + CRUISE

-----

## ELECTRONICS (CONT'D)

|                      |               |
|----------------------|---------------|
| GLIDE SLOPE REC..... | AN/ARN-18     |
| LOCALIZER.....       | (2) AN/ARN-14 |
| LF COMM. REC.....    | (2) BC-348    |
| NAV RADAR.....       | AN/AFS-42     |
| UHF COMMAND.....     | AN/ARC-27     |